Manual amendments

Rev. No.	Page	Description	Date	Signature
1	3, Diagr.9	TN 348/1T	March 92	
2	2, 3, 7, 8, 14, 15, 17, 18, 23, 25, 34	TN 348/3T	Oct. 92	
3	3, Diagr.7	ÄM 500/7/93	June 93	
4	2, 3, 17 Instruction for inspection, questionaire, working instructions No.1 and No.2 for TN 348/4	TN 348/4T	Oct. 94	
5	2, 4, 15, 18, 21, 25	TN 348/9	Oct. 97	
6	2, 24	TN 348/15	Jan. 01	
7	2-4, 7, 17, 21, 22, 35, 35a	Manual revision TN348/20	May 08	

Instructions for continued airworthiness

Content		Page	LBA approved
0	Airworthiness limitations	4	May 08
1.	**System description and adjustment data**		**issued**
1.1	Wing and tailgroup setting data	5	July 90
1.2	Elevator control and trim system	6	" "
		7	May 08
1.3	Rudder control	8	Oct. 92
1.4	Aileron control	9	July 90
		10	" "
1.5	Airbrake control and wheel brake	11	" "
1.6	Undercarriage	12	" "
		13	" "
1.7	Tow hooks	14	Oct. 92
1.8	Sectionnot effective		
1.9	Massbalance and weights of control surface	15	Oct. 97
1.10	Fore and aft play of the wings	16	July 90
2.	**Inspections**		
2.1	Daily inspection	17	May 08
2.2	Regular inspections	17	"
2.3	Inspection after a heavy landing	18	Oct. 97
		19	July 90
		20	" "
2.4	Inspection procedure for increase of service time	21	May 08
		22	May 08
3.	**Maintenance**		
3.1	General maintenance	23	Oct. 92
3.2	Maintenance of the airframe	24	Jan. 01
3.3	Greasing programme	24	" "
3.4	Damage of the airframe	25	Oct. 97
3.5	Hydraulic brake system	25	Oct. 97
4.	**Detailed instructions for assembly and servicing work**		
4.1	Section not effective		
4.2	Replacement of control cables	26	July 90
4.3	Adjustment and servicing of the control circuit	27	" "
4.4	Removal and installation of the undercarriage	28	" "
		29	" "
4.5	Filling and bleeding the hydraulic disc wheel brake	30	" "
4.6	Removal and installation of the under-Carriage (non retractable)	31	" "

Instructions for continued airworthiness

Content		**Page**	**issued**
5.	Centre of gravity measurements-weighing	32 33	July 90 " "
6.	Instruments and accessories list	34 35 35a	Oct. 92 May 08 May 08
7.	List of special tools	36	July 90

Diagrams		**issued**
1	Elevator control circuit, trim	April 90
2	Rudder control circuit	" "
3	Aileron and spoiler control circuits, fuselage side	" "
4	Aileron and spoiler control circuits, wing side	" "
5	Tow hooks	" "
6	section not effective	" "
7	Retractable landing gear, hydraulic wheel brake	June 93
8	Pitot/static system	April 90
9	Placards	March 92
10	Non retractable landing gear	April 90

Enclosures

Equipment list		July 90
5 EP 34 Installation Dräger oxygen system		25.01.90
5 EP 30 Installation ELT		27.02.91
Instruction for inspection	for TN 348/4T	Oct. 94
Questionnaire	for TN 348/4	Oct. 94
Working instruction No. 1	for TN 348/4	Oct. 94
Working instruction No. 2	for TN 348/4	Oct. 94
5V18 Tool for airbrake adjustment		14.10.94

Instructions for continued airworthiness

0 Airworthiness limitations

0.1 Repairs:

Repair damaged wings, fuselage and tail surfaces prior to next flight.
Repairs outside the scope of DG Flugzeugbau DG-500 ELAN TRAINER repair manual and major repairs must be accomplished at a certified repair station or a certified mechanic rated for composite aircraft structure work in accordance with DG Flugzeugbau repair methods.
Use only genuine spare parts

0.2 Life time of the airframe

The maximum allowable operating time for composite sailplanes is 12000 flight hours. Therefore inspection according to sect. 2.4 of this manual has to be executed at 3000 h, 6000 h, 9000 h and every 1000 hours following thereafter.

0.3 Life time of components

a) The fabric straps of the safety harness have to be exchanged after 12 years.

b) The rubber cords in the elevator control system see sect. 1.2.6 and in the wing flap control system see sect. 1.4.6 have to be replaced at least every 5 years.

c) **Other components**
All other components like tow hook, wheels, gas struts, control system parts, bolts, pins etc. have no life time limitation, but should be replaced when worn, damaged or disqualified by excessive corrosion.

0.4 Service time, maintenance documents

Follow the instructions of the respective manufacturer.

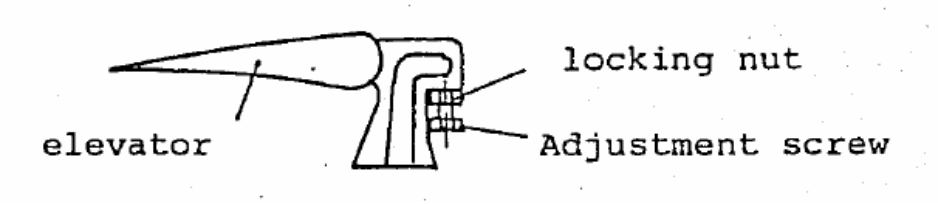
a) Operating Manual for Tow Releases
Series: E 85 Nose Tow Release
Date of Issue: March 1989

Operating Manual for Safety Tow Releases
Series: Europa G 88 Safety Tow Release
Date of Issue: February 1989

b) safety harness: instructions of the manufacturer

c) minimum instrumentation: instructions of the manufacturer

1.2.4 Any free play can be reduced by screwing in the adjustment screw on the automatic connector funnel.

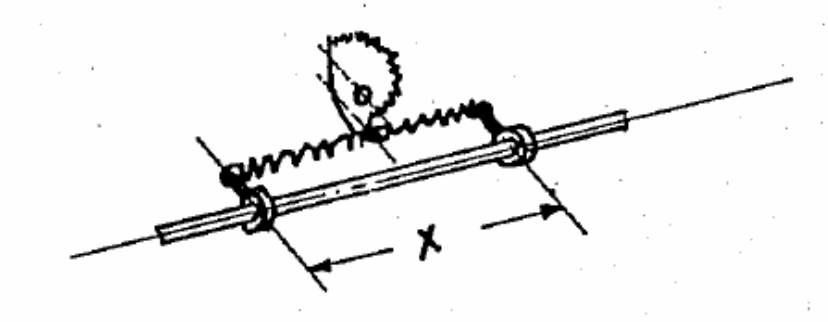


1.2.5 **Trim**

The trim mechanism should be adjusted so that with full forward (nose down) trim the control column is in the maximum forward position.

The tensioning of the trim mechanism springs is adjusted as shown in the sketch. x = 340 mm (13.4 in.)

The springs are located in the rear cockpit on the left hand side.



1.2.6 **Pilot force reducing rubbercord**

The rubber cord produces an elevator stick force in push direction. If the trim efficiency of your DG-500 in push direction is reduced, you have to inspect the rubber cord. The rubber cord is located on the left hand side behind the main bulkhead below the baggage compartment floor. The rubber cord runs from bellcrank 5St19 to a fork at the main bulkhead. The length of the rubber cord when loose should be 500 mm (19.7 in.).
If the cord is longer or worn it must be replaced. The cord must be replaced at least every 6 years.

2. Inspections

2.1 Daily inspection

see flight manual section 4.3

2.2 Regular inspections

A **After 200 flight hours and during the annual inspection**
Check the rudder cables for wear especially around the S tubes on the rudder pedals. Worn rudder cables should be replaced (see section 4.2).
Check the sealing of the rudder (see section 1.3.5).

B **Annual inspection** (and 100hr inspection – only for USA)

- Execute all items of the daily inspection see flight manual section 4.3.
- Inspect all bolted connections and locking devices ie. locknuts, split pins etc.
- Check all metal parts for adequate greasing and rust prevention. (see section 3.3).
- Check the control surface deflections (see sections 1.2 up to 1.4).
- Check the free play in all control circuits (see section 1.2 up to 1.6)
- Check the fore and aft play of the wings (see section 1.10).
- Check the canopy emergency releases according to section 7.15 of the flight manual.
- Check the rubber cords in the control system (see sections, 1.2.6 and 1.7.5.
- Check the thickness of the wheel brake linings (see section 1.6.4).
- Check if the brake fluid has to be exchanged (see section 1.6.4).
- Check the airbrakes according to instructions for inspection for TN348/4T.
- **Tow hooks:** The operating and maintenance instructions for the release mechanisms, see sect. 0.4 of this maintenance manual have to be followed.
- **All-up weight and centre of gravity:** These should be checked at least every 4 years during the annual inspection.

C Section not effective

D **Special inspections**
Tow hook:
After a wheel-up landing, the C.G. tow hook is to be cleaned and to be carefully checked for any damage.
C.G. weighing: After all work which may influence the C.G.

2.4 Inspection procedure for increase of service time

1. General

The results of fatigue tests of wingspan sections have demonstrated that the service time of GFRP/CFRP gliders and motorgliders may be limited to 12000 hours, if for each individual glider (in addition to the obligatory annual inspections) the airworthiness is demonstrated according to a special multi-step inspection program particularly with regard to the service life.

2. Dates

When the glider has reached a service time of 3000 hours, an inspection must be done in accordance with the inspection program mentioned under point 3. If the results of this inspection are positive or if any defects found have been duly repaired, the service time of the glider is extended by another 3000 hours to a total of 6000 hours (first step).

The above inspection program must be repeated when the glider has reached a service time of 6000 hours. If the results of this inspection are positive or if any defects found have been duly repaired, the service time of the glider is extended to 9000 hours (second step).

When the glider has reached a service time of 9000 h the above inspection program must be repeated. If the results of the inspection are still positive, or if any defects found have been duly repaired, the service time may be extended to a total of 10000 hours (third step).

Proceed analogous when reaching 10000 and 11000 hours (4. + 5. step).

3. Ask the manufacturer for the necessary inspection document.
   When you request the inspection document, the following data should be submitted: Model/Type, Registration, Serial Number and the operating hours at which the inspection will be performed. A charge will be made for the inspection document.

4. The inspection must only be done by the manufacturer or by a licensed repair station or inspector.

5. The results of the inspections have to be recorded in an inspection test report wherein comments are required for each inspection instruction. If the inspections are done outside the manufacturer's facilities, a copy of the records must be sent to the manufacturer for his evaluation and information

6.5 **VHF transceiver**

Manufacturer	Type	**Certification No.**
Dittel	FSG-40 S	10.911/45
	FSG-50	10.911/71
	FSG-60 M	10.911/72
	FSG-70,71 M	10.911/81
	FSG-90	10.911/98JTSO
	FSG 2T	LBA.0.10.911/103JTSO
Becker	AR 3201-(1)	10.911/76
	AR 2008/25 (A)	10.911/48
	AR 4201	JTSO-2C37 D, ED-23A
Filser	ATR 720 A	10.911/74
	ATR 720 C	10.911/83
	ATR 600	LBA.0.10.911/106JTSO
	ATR 500	LBA.0.10.911/113JTSO
	ATR 833	EASA.210.0193

or other instruments certified for aircraft use according to TSO or JTSO or ETSO standards may be installed.

Note: Only radios with diameter 58mm (2 ¼ in.) can be installed at the assigned place in the console below the instrument panel.

6.6 **Variometer**

Manufacturer	**Type**	**Certification No.**
Winter 5 St VM5 (dia.58 mm)		TS 10.230/14
	±5 m/sec Ident.No.5451	
	±1000 ft/min ident.No.5452	
	±10 knts Ident.No.5453	
Winter 5 StV5 (dia.80 mm)		TS 10.230/13
	±5 m/sec Ident.No.5251	
	±1000 ft/min Ident.No.5252	
	±10 knts Ident.No.5253	

6.7 **Turn and bank indicator**

Manufacturer	**Type**	**Certification No.**
Apparatebau Gauting	WZ-402/31 12 V	10.241/8

6.8 **Accelerometer**

(for Category A Aerobatics)

Accelerometer capable of retaining max. and min. g-values with markings red radial lines at +7g and -5g.

Manufacturer	**Type**	**Certification No.**
AOA Apparatebau Gauting	BM 470-RL/L	MIL-A-5885 A
Bendix	2" 5V LITE	MS 28025-1
Bendix	3419-5A-A1	MS 28025-1
Burton Manufacturing Co.	B-6	MS 280025-1
INSCO	6610	MS 33638
Kelvin & Hughes Ltd.	KAE 0504K	MS 23009-1
Milhard Engineering Co	ABU-4/A	MS 23009-1
QED/Inc. (ASG)	ABU-4/A	MIL-A-25949
Smiths	KAE 0504/K	MS 23009-1
Falcon Gauge	GMS 10-2	MIL-A-5885 C

6.9 **Instruments which are not part of the minimum equipment:**

Transponders: Transponders certified for aircraft use according to TSO or JTSO or ETSO standards may be installed.

Other instruments and equipment (eg. variometers, gliding computers or flight data recorders):
Instruments and other equipment may be installed if they do not in themselves, or by their effect upon the sailplane, constitute a hazard to safe operation.

Caution: If additional instruments or equipment are to be installed after production of the glider, it must be assured that they will be installed in the places provided by the design. If installed in other places it must be assured that they are secured safely.
Electrical instruments and equipment must be connected via appropriately rated fuses, the power consumption of each single part should not exceed 3A.

After installation raise a new weight and balance report.